Amendments to the Claims:

Please cancel claims 1 to 12 as presented in the underlying International Application No. PCT/DE2004/002192 without prejudice.

Please add new claims as indicated in the listing of claims below.

This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims:

Claims 1 to 12 (cancelled).

Claim 13 (new): A turbine engine, comprising:

a compression stage including a rotor and a plurality of turbine blades positioned side by side in a circumferential direction of the rotor, each turbine blade having a blade root, each turbine blade being securable via the blade root in a retainer groove which extends in the circumferential direction of the rotor, each turbine blade being insertable with its blade roots into the retainer groove via a filling groove, the width of the blade roots and the width of the filling groove being greater in the circumferential direction than one-half of the width of a desired, nominal blade pitch;

wherein the plurality of turbine blades include a first plurality of turbine blades and a second plurality of turbine blades, the first plurality of turbine blades having the desired, nominal blade pitch and the second plurality of turbine blades having a blade pitch greater than the desired nominal blade pitch, the second plurality of turbine blades being in an area of the filling groove, the first plurality being greater in number than the second plurality.

Claim 14 (new): The turbine engine as recited in claim 13, comprising a plurality of compression stages.

Claim 15 (new): The turbine engine as recited in claim 13, comprising a plurality of filling

grooves.

Claim 16 (new): The turbine engine as recited in Claim 15, wherein the plurality of filling grooves are uniformly distributed over the circumference of the rotor, the width of each filling groove being greater in the circumferential direction than one-half of the width of a desired, nominal blade pitch, and, for each filling groove, the first plurality of turbine blades have the desired, nominal blade pitch and the second plurality of turbine blades have a blade pitch greater than the desired nominal blade pitch, the second plurality of turbine blades being in an area of the filling groove, the first plurality being greater in number than the second plurality.

Claim 17 (new): The turbine engine as recited in Claim 13, comprising two filling grooves facing one another diametrically.

Claim 18 (new): The turbine engine as recited in Claim 13, wherein three turbine blades having the desired, nominal blade pitch can be replaced in the area of the filling groove by two turbine blades having the increased blade pitch.

Claim 19 (new): The turbine engine as recited in Claim 13, wherein the second plurality of turbine blades having an increased blade pitch have an enlarged platform area in the circumferential direction compared to the first plurality of turbine blades having the desired, nominal blade pitch.

Claim 20 (new): The turbine engine as recited in Claim 13, wherein the width of the filling groove in the circumferential direction corresponds approximately to one-half of the width of the increased blade pitch.

Claim 21 (new): A rotor for a compression stage of a turbine engine, comprising:

a plurality of turbine blades positioned side by side in a circumferential direction of the rotor, each turbine blade having a blade root, each turbine blade being securable via the blade root in a retainer groove which extends in the circumferential direction of the rotor, each turbine

blade being insertable with its blade roots into the retainer groove via a filling groove, the width of the blade roots and the width of the filling groove being greater in the circumferential direction than one-half of the width of a desired, nominal blade pitch;

wherein the plurality of turbine blades include a first plurality of turbine blades and a second plurality of turbine blades, the first plurality of turbine blades having the desired, nominal blade pitch and the second plurality of turbine blades having a blade pitch greater than the desired nominal blade pitch, the second plurality of turbine blades being in an area of the filling groove, the first plurality being greater in number than the second plurality.

Claim 22 (new): The rotor as recited in claim 21, comprising a plurality of compression stages.

Claim 23 (new): The rotor as recited in claim 21, comprising a plurality of filling grooves.

Claim 24 (new): The rotor as recited in Claim 23, wherein
the plurality of filling grooves are uniformly distributed over the circumference of the rotor, the width of each filling groove being greater in the circumferential direction than one-half of the width of a desired, nominal blade pitch, and, for each filling groove, the first plurality of turbine blades have the desired, nominal blade pitch and the second plurality of turbine blades have a blade pitch greater than the desired nominal blade pitch, the second plurality of turbine blades being in an area of the filling groove, the first plurality being greater in number than the second plurality.

Claim 25 (new): The rotor as recited in Claim 21, comprising two filling grooves facing one another diametrically.

Claim 26 (new): The rotor as recited in Claim 21, wherein three turbine blades having the desired, nominal blade pitch can be replaced in the area of the filling groove by two turbine blades having the increased blade pitch.

Claim 27 (new): The rotor as recited in Claim 21, wherein the second plurality of turbine

blades having an increased blade pitch have an enlarged platform area in the circumferential direction compared to the first plurality of turbine blades having the desired, nominal blade pitch.

Claim 28 (new): The rotor as recited in Claim 21, wherein the width of the filling groove in the circumferential direction corresponds approximately to one-half of the width of the increased blade pitch.